

Review

# Sustainable Aviation Electrification: A Comprehensive Review of Electric Propulsion System Architectures, Energy Management, and Control

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**Abstract:** The civil aviation sector plays an increasingly significant role in transportation sustainability in the environmental, economic, and social dimensions. Driven by the concerns of sustainability in the aviation sector, more electrified aircraft propulsion technologies have emerged and form a very promising approach to future sustainable and decarbonized aviation. This review paper aims to provide a comprehensive and broad-scope survey of the recent progress and development trends in sustainable aviation electrification. Firstly, the architectures of electrified aircraft propulsion are presented with a detailed analysis of the benefits, challenges, and studies/applications to date. Then, the challenges and technical barriers of electrified aircraft propulsion control system design are discussed, followed by a summary of the control methods frequently used in aircraft propulsion systems. Next, the mainstream energy management strategies are investigated and further utilized to minimize the block fuel burn, emissions, and economic cost. Finally, an overview of the development trends of aviation electrification is provided.

**Keywords:** environmental sustainability; aviation electrification; system architectures; control system design; setpoint tracking control; transient control; safety constraints; energy management strategy; fuel economy; emissions



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## 1. Introduction

The global aircraft fleet consumes over 7% of oil products (276 million tons of jet fuel) and produces 2.7% of energy-consumption-related CO<sub>2</sub> emissions due to in-flight combustion. Furthermore, the global aviation demand is anticipated to grow at around 4.8% annually [1]. By the year 2050, a greater than 60% increase in global commercial air travel seat miles and a 38% increase in energy use are projected by the U.S. Energy Information Administration, with corresponding CO<sub>2</sub> emissions projections of 209 million metric tons CO<sub>2</sub>e [2]. Thus, it is critical to reduce the environmental footprint of the aviation sector, and the civil aviation sector plays an increasingly significant role in transportation sustainability in the environmental, economic, and social dimensions. The major organizations and research councils have all published pathways for sustainable aviation decarbonization to reduce aviation-related pollutant emissions. For example, the NASA 'N+3' strategic implementation plan aims for −75% NO<sub>x</sub> emissions, −70% fuel burn, and −55 dB noise at the airport boundary in the year 2035 [3], and Flightpath 2050 targets a 75% reduction in CO<sub>2</sub> emissions per passenger kilometer and a 90% reduction in NO<sub>x</sub> emissions relative to the technology level of the year 2000 [4]. However, the potential efficiency improvements achieved by conventional technology progress in airframes, structure, propulsion, and air-traffic management are already reaching a plateau. Driven by the concerns of environmental sustainability in the aviation sector, electrified aircraft propulsion technologies have

emerged and have been identified as the most promising approach to realize sustainable and decarbonized aviation.

Many historical surveys have been published to provide a mature insight into electrified aircraft propulsion systems. For example, the existing potential and forecasted progress of aircraft hybrid electric architectures and their impact on the environment are surveyed in [5]. The recent progress, opportunities, and challenges of conceptual design and the technical advances of electrical system components are presented in [6], along with a discussion of the aircraft-level effects of electrification and the emerging problem of thermal management. Furthermore, the research work and development trends of conceptual design, preliminary sizing, and multidisciplinary optimization frameworks are comprehensively reviewed in [1], which also includes original and detailed assessments of regulations, certifications, and infrastructure. Furthermore, a life-cycle-oriented sustainability assessment of emerging aircraft technologies is investigated in [7] to identify these hotspots and problem shifts and provide further guidance for actions aimed at sustainable aviation development at an early stage. Additionally, technical advances in aviation electrification are explored to enhance strategy research and development investment analysis in [8].

This paper aims to provide a comprehensive and broad-scope survey of the recent progress and development trends in aviation electrification. In Section 2, the major architectures of electrified aircraft propulsion systems are presented from the perspectives of concepts, benefits, challenges, and studies/applications to date, and we further identify the performance requirements of the electrical system components. Section 3 discusses the challenges and technical barriers of the control system design of electrified aircraft propulsion systems and then summarizes the mainstream control techniques for aircraft propulsion systems. Next, the research gap on hybrid electric aircraft energy management strategies is discussed in Section 4, followed by a review of the recent progress in energy management strategy research. Finally, Section 5 presents an overview of the development trends of aviation electrification.

## 2. Electrified Aircraft Propulsion System Architectures

Electric propulsion architectures are generally classified as: all-electric, series hybrid electric, parallel hybrid electric, and turboelectric [9]. In this section, each architecture will be introduced individually from the perspectives of concepts, benefits, challenges, and studies/applications to date.

### 2.1. All-Electric Architecture

In all-electric propulsion architecture, batteries are the only energy source for the propulsion systems, as shown in Figure 1. The all-electric configuration is a high-efficiency energy conversion system and is the only candidate that has the potential to achieve zero emissions [10].

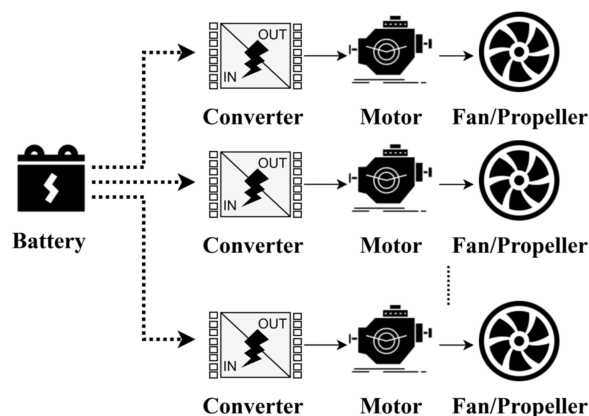


Figure 1. All-electric architecture [9].

All-electric aircraft are identified as a feasible and cost-competitive approach to reduce the environmental impacts of short-haul flights, including vertical take-off and landing aircraft (eVTOLs) [11,12], commuter aircraft [13], rotorcraft [14,15], and general aviation airplanes. The key applications and studies to date are summarized in Table 1. The potential benefits that can be exploited from the global aircraft fleet are critically dependant on battery-pack technical advances [16]. All-electric aircraft with a battery-pack specific energy of 800 Wh/kg enable a flight mission range of up to 600 nmi, which can cover half of all aircraft departures. At this stage, NOx emissions at airport areas can be significantly mitigated by 40%, and block fuel burn and direct CO<sub>2</sub> emissions can be reduced by 15%. To reach cost effectiveness with conventional aircraft, carbon taxes and batteries should be below USD 100 per kWh [17]. To enable a flight mission of 900 nmi with an all-electric propulsion system, the battery specific energy is required to reach 1200 Wh/kg [18].

**Table 1.** Applications/studies/conceptual design of all-electric aircraft to date.

Study	Category	Technical Specification	Key Findings
Airbus E-Fan [19]	Two-seat electric aircraft for pilot training. Endurance: 60 min; first flight year: 2014	Battery: Li-ion, 207 Wh/kg EM power: 30 kW	Two electric motors via eight-blade ducted fans, each producing a thrust of 0.75 kN.
Magnus eFusion [20]	Two-seat training aircraft with light aerobatic capabilities, serving as flying testbed for the sub-100 kW electric propulsion system. First flight year: 2016.	Propulsion system: RRP70D EM: continuous power 70 kW	
Siemens Extra 330 LE [20]	Two-seat aerobatic aircraft as a flying testbed for motors in the class of 0.25 to 0.5 MW. First flight year: 2016	Propulsion system: RRP 260D EM power: 260 kW EM power density: 5.2 kW/kg	
Airbus Vahana [21]	Single-seat eVTOL. Range: 50 km; first flight year: 2019	EM power: 8 × 45 kW Batteries: 38 kWh	All-electric propulsion system for a tilt-wing aircraft configuration; flight testing culmination time: 12 h, totaling 500 nmi.
Boeing Aurora [22]	2 PAX eVTOL. Target YEIS: 2020		21 ft long with a 36 ft wingspan, 12 independent lift fans.
Rolls-Royce/YASA ACCEL Project [23]	Single-passenger light sport and training aircraft. Range: 200 miles; first flight year: 2020		Developing the world's fastest electric aircraft, expected to reach 480 km/h.
NASA X-57 Maxwell [24]	Two-seat Tecnam P2006T general aviation aircraft. Maximum operational altitude: 14,000 ft; cruise speed: 172 mph; first flight year: 2020	Aircraft weight: 3000 lb Batteries: Li-ion, 69.1 kWh Power distribution: 460 V	14 motors and propellers (two large cruise motors and propellers and 12 small high-lift motors and propellers located across the wing to increase airflow).
EVATION ALICE [25]	9 PAX commuter aircraft. Range: 440 nmi; maximum cruise speed: 250 kts; maximum payload: 2500 lbs; target YEIS: 2021	MTOW: 16,500 lbs Propulsion: magni650 Max Power: 2 × 640 kW	
Airbus CityAirbus [26]	Four-seat eVTOL. Range: 80 km; cruise speed: 120 km/h; target YEIS: 2023	MTOW: 2200 kg Propulsion system: RRP200D EM power: 8 × 200 kW EM torque density: 30 Nm/kg	A fixed wing with V-shaped tail; eight electric-powered propellers with distributed propulsion system.
Wright Electric/Easy Jet [27]	100 PAX large commercial aircraft on the platform of BAe 146. Endurance: 1 h; target YEIS: 2026	EM power: 2 MW EM SP: 10 kW/kg Inverter power: 2 MW Inverter frequency: 300 kHz Inverter volume density: 20 kW/L	10 × 2 MW motors totaling 20 MW, as powerful as an A320 Airbus aircraft.
Rolls-Royce/Siemens CleanSky 2 ELICA [28,29]	19 PAX commuter aircraft. Target YEIS: 2060	Battery SE: 500 Wh/kg Battery SP: 1 kW/kg EM SP: 7.7 kW/kg	All-electric concept with 16 distributed propellers for design range of 200 nmi and 400 nmi.

\* EM: electric machine, MTOW: maximum take-off weight, SE: specific energy, SP: specific power, PAX: passenger, YEIS: year of entry into service.

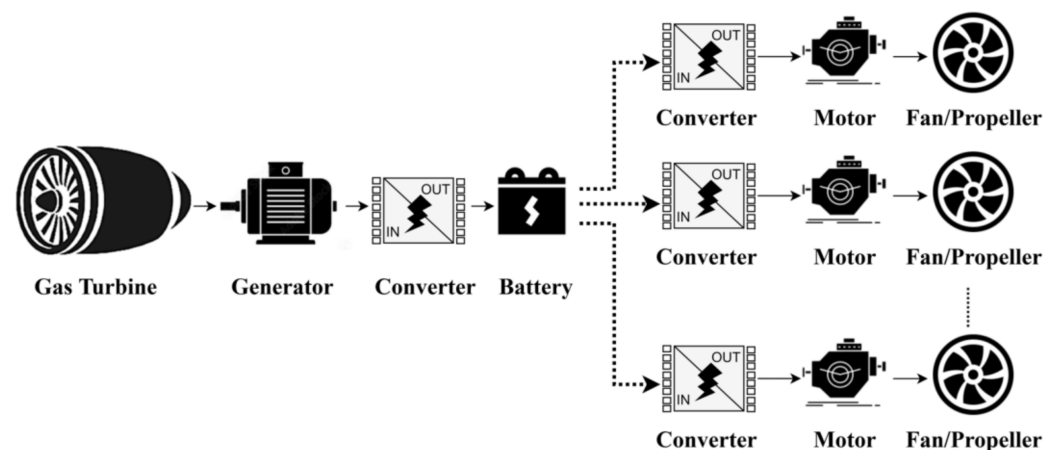
The Committee on Propulsion and Energy Systems to Reduce Commercial Aviation Carbon Emissions identified the electrical system component performance requirements for all-electric propulsion systems, as shown in Table 2 [10].

**Table 2.** Electrical system component performance requirements for all-electric propulsion systems.

Aviation Sector	EM Power Capacity	EM Specific Power	Battery Specific Energy
General aviation	Motor: <1 MW	Motor: >6.5 kW/kg	>400 Wh/kg
Regional/single-aisle	Motor: 1–11 MW	Motor: >6.5 kW/kg	>1800 Wh/kg
Twin-aisle	Not feasible	Not feasible	Not feasible

## 2.2. Series Hybrid Electric Architecture

The series hybrid electric architecture is presented in Figure 2. It is normally made compatible with distributed propulsion by using multiple relatively small electric motors and propellers/fans [10]. In series configurations, the fans/propellers are driven by electric motors, and the electrical power is either from a gas-turbine-driven generator or from a battery energy storage source. Because of the electrical power conversion and the transmission system, the operation of the gas turbine can be decoupled from fans/propellers, which enables the gas turbine to operate at its maximum efficiency. However, in series-hybrid electric propulsion systems, the electric motors should provide all of the propulsive power. Consequently, the electric motors need to be the right size to achieve their maximum power, which will increase the mass and volume of the propulsion system [30]. A summary of series hybrid electric architecture studies is provided in Table 3.

**Figure 2.** Series hybrid electric architecture [9].**Table 3.** Applications/studies/conceptual design of series hybrid electric aircraft to date.

Study	Category	Technical Specification	Key Findings
Zunum Aero [31]	Seat capacity: 12 economy, 9 premium, 6 executives; max payload: 2500 lbs; range: 700 nmi; target YEIS: 2020	Propulsion system: series hybrid with range extender Max power: 1 MW Turbogenerator: 500 kW	Emissions: 0.0 to 0.3 lbs CO <sub>2</sub> /ASM Operating cost: 8 cents/seat mile, USD 250 per hour.
XTI TriFan 600 [32]	6-seat fixed-wing aircraft with VTOL. Range: 600 nmi in VTOL, 900 nmi for conventional take-off and landing; YEIS: 2024	Propulsion system: a turboshaft engine driving 3 generators for electrical energy generation, powering motors which are mechanically connected to propellers	Three ducted fans, hybrid energy system (hydrogen fuel cell, sustainable aviation fuel compatible).
Airbus E-Fan X [33]	100-seat regional jet. Payload: 6650 kg; YEIS: 2030	Motor power: 2 MW Generator power: 2.5 MW EM power density: 10 kW/kg Power distribution: 3 kV DC	One of the four jet engines (AE2100) was replaced by a 2 MW electric motor.



### 2.3. Parallel Hybrid Electric Architecture

In a parallel hybrid electric architecture, as shown in Figure 3, both the electric motor and the conventional gas turbine are mounted on a shaft to drive the fans/propellers. The electric power on-take can be used to boost either the low-pressure shaft, known as Mechanically Integrated Parallel Hybrid architecture (MIPH), or the high-pressure shaft, known as Cycle-Integrated Parallel Hybrid architecture (CIPH) [34]. Additionally, the electric motor is backed up with a battery energy storage system that can be charged onboard or on the ground.

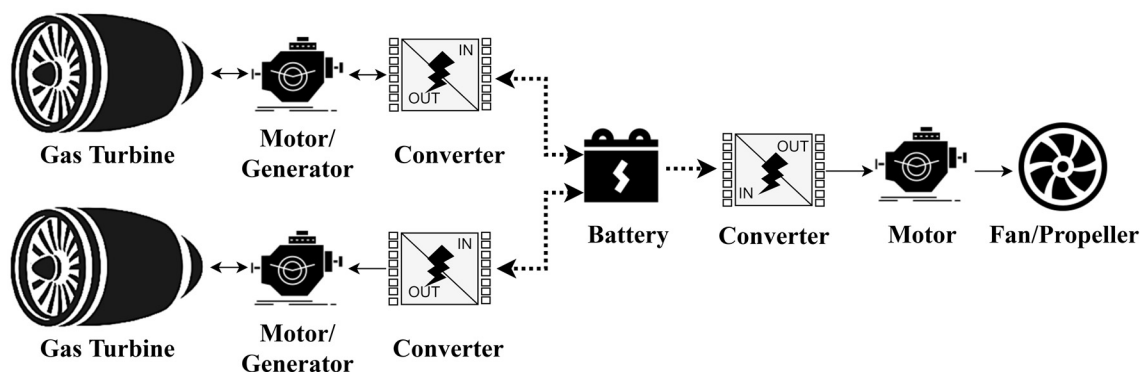


Figure 3. Parallel hybrid electric architecture [9].

The parallel configuration requires fewer electrical components compared with the serial hybrid configuration, so it has the advantage of weight saving [35]. However, the operation and control in the parallel electric configuration are relatively complicated due to the mechanical coupling relations of the propulsive device, gas turbine, and electrical system. Furthermore, the parallel hybrid electric operation may deteriorate the performance of the turbomachinery components. For example, in MIPH arrangements, under a fixed thrust demand, hybridization operation reduces the power demand from the engine core, and the gas turbine operation of the overall pressure ratio, pressure build-up, and core/bypass mass flow is noticeably affected due to the components' rematching effects, as shown in Figure 4 [36]. To provide a lower shaft power, the rotational speed of the high-pressure shaft is accordingly reduced, which causes a flow mismatch between the two spools [37]. This will eventually lead to a reduced surge margin in the low-pressure compressor and significantly increase the fan rotational speed [34]. For CIPH configurations, the low-pressure compressor is more likely to choke because of the components' rematching caused by hybridization, and the high-pressure shaft compressor moves towards a high rotational speed [34]. Thus, the operation of the parallel hybrid configuration is complicated, because it should avoid potential fluttering (stalling or choking) and ensure mechanical integrity.

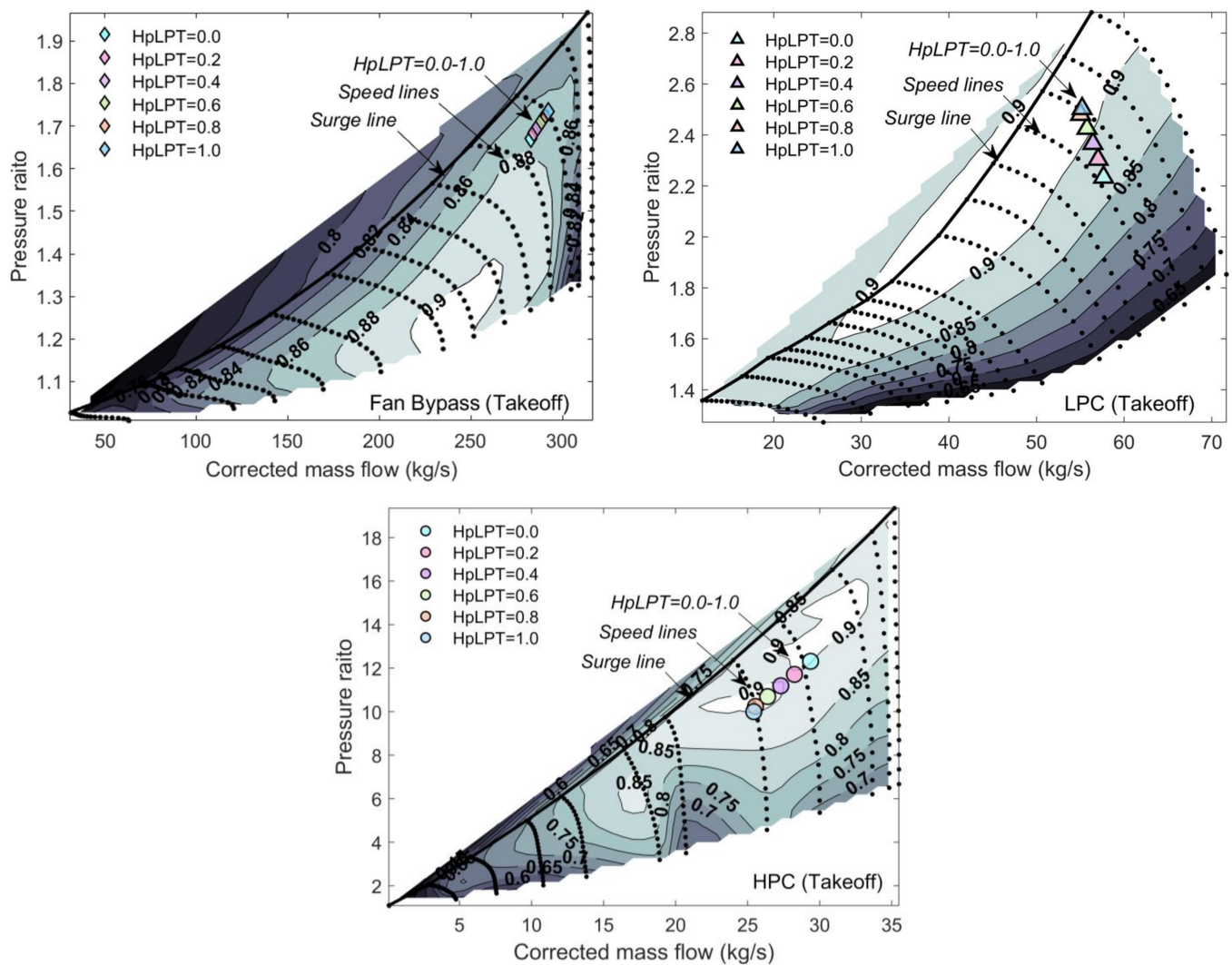


Figure 4. Parallel hybrid electric operation in take-off conditions [36].

The applications of and ongoing research on parallel hybrid electric architecture are summarized in Table 4.

Table 4. Applications/studies/conceptual design of parallel hybrid electric aircraft to date.

Study	Category	Technical Specification	Key Findings
NASA Boeing SUGAR Volt [37,38]	150 PAX single-aisle aircraft. Range: 900 nmi	Battery SE: 750 Wh/kg EM SP: 3–5 kW/kg EM efficiency: 93% EM power capacity: 1.3 MW (balanced), 5.3 MW (core shutdown)	Transonic truss-braced wing, ‘balanced’ version: reduces fuel burn by 60% and energy use by 54%. ‘Core shutdown’ version: EM for cruise on 100% electric power, reduces fuel burn by 64% and energy use by 46%.
NASA UTRC hGTF [38,39]	150 PAX single-aisle aircraft. Range: 900 nmi	EM power capacity: 2.1 MW Battery SE: 1000 Wh/kg	Optimized geared turbofan engine for cruise, electric power boosting for take-off and climb; 7–9% block fuel burn reduction and 3–5% energy saving.
NASA R-R LibertyWorks EVE [39,40]	150 PAX single-aisle aircraft, exploring mission optimization using battery power for taxiing, idle decent, and take-off power augmentation	EM power capacity: 1 MW–2.6 MW	28% fuel burn reduction for 900 nmi mission, 10% energy saving for 500 nmi mission, 18% reduction in total fleet fuel usage.

Table 4. Cont.

Study	Category	Technical Specification	Key Findings
Horizon 2020 H3PS (High-Power High-Scalability Aircraft Hybrid Powertrain) [40]	4-seat general aviation aircraft developed on the platform of Tecnam P2010	Engine: Rotax 915 (141 hp) EM power capacity: 30 kW (thrust booster motor during take-off and climb, operating as a generator to recharge batteries during cruise)	TECNAM: airframe and system integration, BRP-ROTAX: design and integration of combustion engine and e-motor, ROLLS-ROYCE: e-motor and power storage.
Clean Sky 2 NOVAIR project [41,42]	150 PAX single-aisle large passenger aircraft (retrofitted on A320 NEO). Range: 800 nmi	Technology level: 2040+ Battery efficiency: 95.0% Battery SE: 1 kWh/kg EM efficiency: 98.0% EM SP: 15.0 kW/kg	With downscaled, more efficient turbofan engine core, potential trip fuel reduction is about 14%.

The technology levels of the electrical system components required by parallel hybrid electric architecture are investigated in [10] and presented in Table 5. The potential applications of all-electric and parallel hybrid electric propulsion systems are largely dependent on the technical advances in battery energy storage systems. A regional jet or single-aisle aircraft is feasible with a battery specific energy of over 800 Wh/kg for a parallel architecture. However, the current state-of-the-art battery specific energy is only 200–250 Wh/kg [10].

Table 5. Electrical system component performance requirements for parallel hybrid electric propulsion systems.

Aviation Sector	EM Power Capacity	EM Specific Power	Battery Specific Energy
General aviation	Motor: <1 MW	Motor: >3 kW/kg	>250 Wh/kg
Regional/single-aisle	Motor: 1–6 MW	Motor: >3 kW/kg	>800 Wh/kg
Twin-aisle	Not studied	Not studied	Not studied

#### 2.4. Turboelectric Architecture

In the turboelectric configuration as shown in Figure 5, the gas turbine drives the generators to provide electricity. The electricity is subsequently transmitted to power inverters, individual direct current (DC) electric motors, and eventually drives the individual distributed fans/propellers. The turboelectric configuration does not include electrical energy storage; it stores all energy in jet fuel and converts part or all of it to electricity. The key studies on turboelectric research are listed in Table 6.

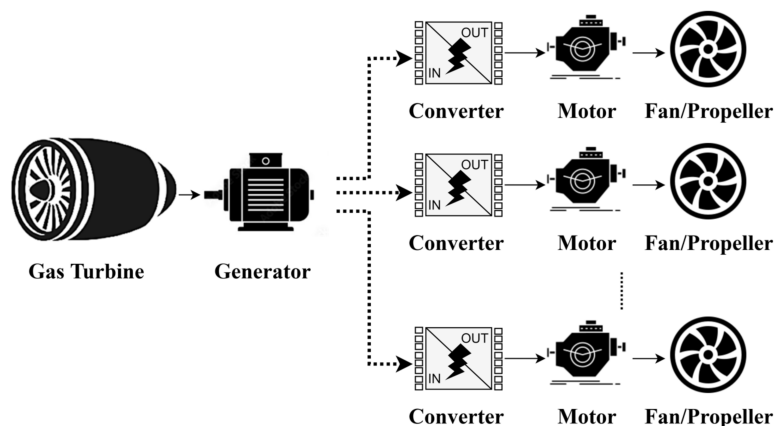


Figure 5. Turboelectric architecture [9].

**Table 6.** Applications/studies/conceptual design of turboelectric aircraft to date.

Study	Category	Technical Specification	Key Findings
NASA N3-X [38,43]	300 PAX hybrid wing body with fully distributed propulsion, 16-aft motor-driven fans. Range: 7500 nmi	Superconducting electric machines and power distribution. Power distribution: 7500 V Fully distributed power: 50 MW	70% fuel burn reduction compared to Boeing 777–200 LR.
Empirical Systems Aerospace ECO-150 R [38,44,45]	150 PAX regional jet, fully distributed propulsion system with a split-wing concept. Maximum payload range: 1500 nmi	Superconducting electrical machines cooled with liquid hydrogen	16-wing motor-driven fans.
NASA STARC-ABL [38,46]	154 PAX single-aisle aircraft with tube and wing airframe. Range: 900 nmi; target YEIS: 2035	EM SP: 8 hp/lb EM efficiency: 96% Inverter SP: 10 hp/lb Inverter efficiency: 98% Power distribution: 1000 V Motor power capacity: 2.6 MW	12% reduction in start of cruise TSFC, 9% reduction in economic mission block fuel, 15% reduction in design mission block fuel.
Boeing SUGAR Freeze [38,47]	154 PAX single-aisle aircraft using a truss-braced wing combined with boundary-layer ingestion. Range: 900 nmi	Solid oxide fuel cell, superconducting motor, cryogenic power management system	56% reduction in fuel burn.

Turboelectric architecture with distributed propulsion has been identified as the top priority approach for developing advanced electric propulsion technologies that could be implemented in the next c. 10–30 years for aviation decarbonization, because it does not rely on the technical advances of battery energy storage. Furthermore, the performance requirements of the electrical system components can be developed with smaller advances beyond the state-of-the-art technology, as shown in Table 7 [10].

**Table 7.** Electrical system component performance requirements for turboelectric propulsion systems.

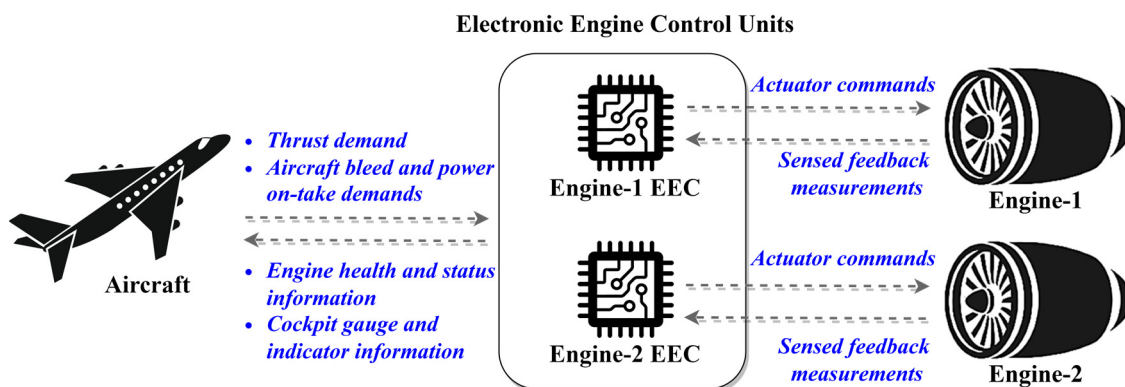
Aviation Sector	EM Power Capacity	EM Specific Power	Battery Specific Energy
General aviation	Motor and generator: <1 MW	Motor and generator: >6.5 kW/kg	NA
Regional/single-aisle	Motor: 1.5–3 MW Generator: 1–11 MW	Motor and generator: >6.5 kW/kg	NA
Twin-aisle	Motor: 4 MW Generator: 30 MW	Motor and generator: >10 kW/kg	NA

### 3. Electrified Aircraft Propulsion System Dynamic Modeling and Control System Design

In this section, the challenges and technical barriers of the control system design for advanced electrified aircraft propulsion (EAP) systems are presented. Then, the control methods for integrated power and propulsion systems are introduced with the latest research.

In a conventional gas-turbine aero-engine control architecture, as shown in Figure 6, the electronic engine control (EEC) units calculate the control commands according to the received requests of thrust demands, aircraft power, and bleed offtake demands and the engine-sensed feedback measurements. Then, the control commands are sent to the actuator of the engine-fuel metering valve. Herein, the fuel flow rate is the primary control variable to control the engine thrust output. As the thrust cannot be measured directly, a closed-loop fuel flow controller is normally designed based on a correlated feedback measurement of the fan speed or engine pressure ratio [48]. Furthermore, the actuators of the variable-area fan nozzle (VAFN) and variable bleed valve (VBV) are open-loop scheduled to improve engine operability. The control system design of aircraft propulsion systems is highly challenging and complex due to nonlinearities, performance constraints, and physical and safety limits [49]. The operation of a gas turbine is an extremely complex nonequilibrium and nonlinear aerothermodynamic process, and the model accuracy is

expected to capture the interaction accurately enough for control design but without requiring excessive computational efforts [50,51]. Additionally, the aircraft operates in a wide flight envelope over an extreme range of environmental conditions, defined by altitude, Mach number, and temperature [52]. The control system of a gas turbine aero-engine is not only responsible for maintaining thrust at a specified level for a given throttle, guaranteeing a fast and smooth transition operation between demanded thrust levels, but must also take into consideration the safety, reliability, maintainability, and environmental impact [53]. The control design is a highly demanding task necessitating the fulfilment of a set of performance and safety requirements including the steady state accuracy, transient accuracy, disturbance rejection, stability, stall margin, structural integrity, and engine degradation. In summary, aircraft propulsion control systems can be classified as in Table 8. The control system design procedure is normally as follows: the flight envelope is divided into several regions. In each region, the linear model is obtained for the flight conditions, so that the controllers are designed for each linear model using control theories. Finally, the full flight envelope control is achieved via gain scheduling [54].



**Figure 6.** Control architecture of conventional gas-turbine propulsion system [48].

**Table 8.** Aircraft propulsion control systems and performance requirements [54].

Control System	Performance Requirements
Setpoint control	Regulate the gas turbine performance near a desired operating condition, e.g., idle, take-off, cruise
Transient control	Transient operation (performance variables change with time)
Limit protection	Physical limits: shaft speed, turbine blade maximum temperature, maximum combustion pressure, surge/stall of compressor

Future electrified propulsion systems will pose further significant challenges because of new configurations, structures, and electric components, presenting much more complex problems with significantly increased numbers of control system parameters to design and tune [55]. In the EAP control architecture, as shown in Figure 7, the supervisory controller is the communication interface between the aircraft and the propulsion system and is also responsible for coordinating the operation of both subsystems, the conventional gas turbine and the electrical powertrain. The supervisory control strategies are proposed to minimize fuel/energy consumption, lessen the challenges of thermal management, achieve dynamic control to ensure engine operability during transients, and maintain overall operating limits [48]. In the bottom layer, EEC units control the gas turbine's operation, and the electronic component controller (including the motor control unit, generator control unit, power electronic control unit, and battery management system) regulates the operation of the electrical powertrain.



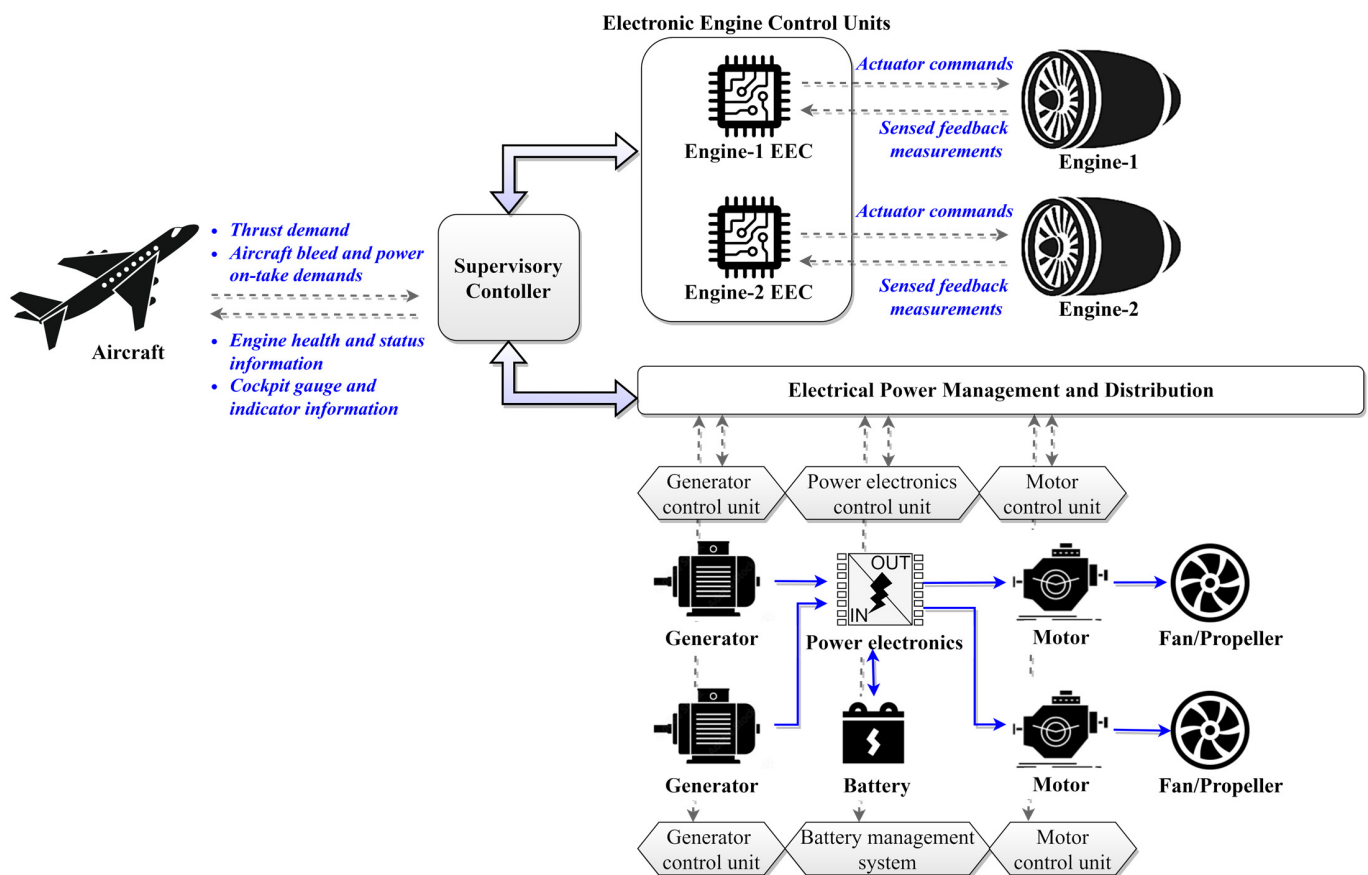
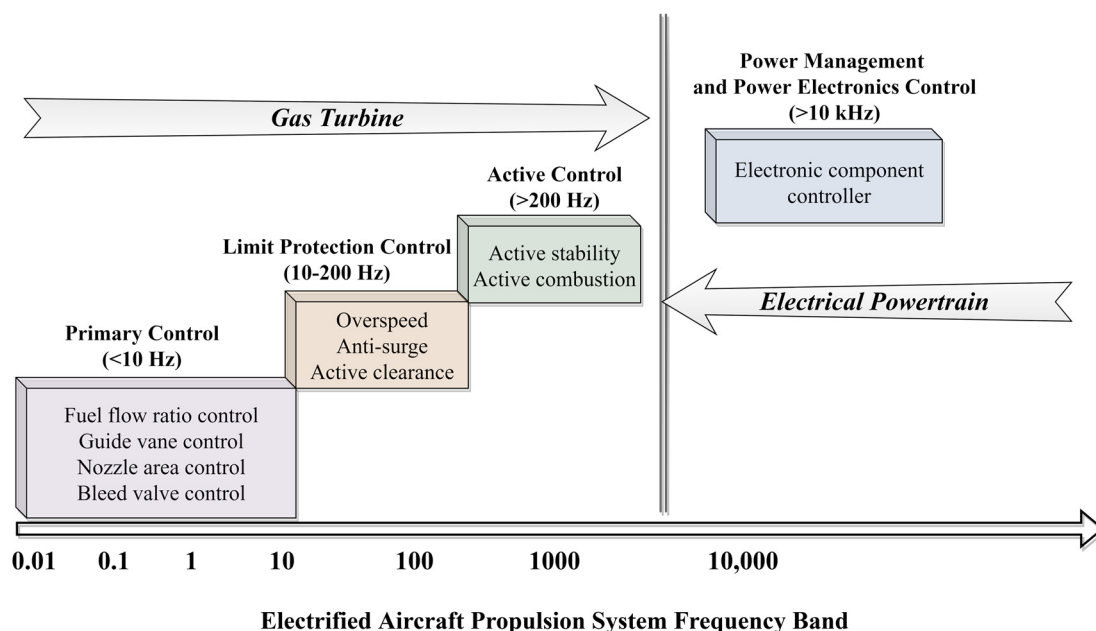


Figure 7. Control architecture of electrified propulsion system [48].

Apart from the existing technical barriers of conventional gas-turbine aero-engine control, the EAP control system design brings with it more challenges and concerns. Firstly, the coupling and interaction between the gas turbine and electrical powertrain should be considered, which involve mechanical, hydraulic, thermal, electrical, and electronic elements. The interaction can result in a loss of setpoint tracking and safety constraint violations, and the power on-take provided by electric motors could compromise the gas turbine operability [56]. Taking MPH architecture as an example, the gas turbine operation in terms of the overall pressure ratio, pressure build-up, and core/bypass mass flow is noticeably affected due to the electrical power on-take and component rematching effects. The electrical power on-take will cause a decrease in the resistive torque load on the engine shaft, significantly increase the fan rotational speed, and eventually lead to a reduced surge margin in low-pressure compressors [34]. Another challenge is the negative impedance instability behavior in the power load regulation of the electrical powertrain subsystem, where an increase in voltage across the device's terminals results in a decrease in the electric current, and vice versa [57]. Along with the low inertia of the electrical powertrain subsystem, the negative impedance instability is very likely to destabilize the voltage buses [58]. Furthermore, the dynamics of a conventional gas turbine subsystem and electrical powertrain subsystem have differences in terms of their time scales; specifically, the voltage dynamics of the electrical powertrain have time constants in the order of milliseconds, while the gas turbine's shaft rotational speeds have time constants in the order of seconds [59]. This phenomenon further complicates the control system design of the integrated EAP system, and the local controllers of subsystems are designed for different update rates, which limits the opportunities for communication. The frequency band of interest for EAP control systems is presented in Figure 8. Based on the dynamic response gap between the engine and electrical powertrain, the engine response lag can be potentially compensated by the motor torque to reduce the shaft torque fluctuation [60]. In

addition, more failures may occur in the EAP system due to the diversity of components and the coupled operation. Thus, fault detection and tolerance, or system reconfiguration in response to faults, are expected to be embedded within the control system design [48].



**Figure 8.** Frequency band of EAP control system [48,54].

Table 9 outlines the main control methodologies for aircraft propulsion systems, including both the conventional propulsion system and the electrified aircraft propulsion system. However, the control methodologies for electrified aircraft propulsion systems have not been universally investigated. Each control method will be introduced individually, highlighting the benefits and limitations of its application. Additionally, the distinct differences between classical control theory and modern control theory are presented in Table 9.

### 3.1. Classical Control

The most common example of a classical control method is the proportional integral derivative (PID) controller, which is widely used in practice due to its simple implementation and relative efficiency [61]. However, PID controllers can only be used for single-input–single-output (SISO) systems. Thus, the application of a PID controller in a multivariable aircraft propulsion system relies on system dynamics decoupling to produce multiple SISO systems that can be handled directly, but this approach will limit the allowable controller gains for system stability [62]. Another approach is a control structure consisting of a cascaded series of control loops with a supervisory multivariable controller that will coordinate control loops and generate output setpoints as references for low-level actuator PID controllers, which is well-known as a hierarchical control structure [63]. Furthermore, the classical control theory cannot be applied to nonlinear systems directly; thus, the system linearization at predefined equilibrium points is required before controller design. Additionally, min–max limiting logic is a necessity for system safety constraint handling, e.g., maximum speed limits, combustor pressure limits, and ratio unit limits [49].

The basic design process and control architecture of a commercial turbofan engine is presented in [49]; it contains an engine pressure ratio and fan speed setpoint controller and limit regulators to constrain the transient to guarantee safe operation, including a rotor speed limiter, a combustor pressure limiter, an acceleration schedule, and a ratio unit limiter. These are all implemented by the PI controller with integral wind-up protection, and a min–max strategy is applied to determine the control signal sent to the fuel metering valve. Furthermore, the aircraft propulsion system operates over a wide range of flight

conditions, which adds significant complexity to the control system design. This challenge is handled by using gain scheduling with interpolation, to break a large, complex, nonlinear system into a series of range-limited linear subsystems and tune the controller at each breakpoint. At each time step, the controller gains are obtained by linear interpolation between the nearest defined subsystems [52]. On this basis, meta-heuristic global optimization algorithms, namely the competent genetic algorithm, linkage-learning genetic algorithm, and aggregative gradient-based algorithm, have been respectively applied to solve the engine controller gain-tuning problem for a turbojet gas turbine, with the global optimization objectives of response time, fuel consumption, maneuverability, and safety considerations [55,64]. Additionally, a cascade ensemble radial basis function (RBF)-based optimization algorithm is also proposed for global optimal control gain scheduling for aero-engine transient processes [65]. Fuzzy gain scheduling is proposed for the PID controller tuning, which is achieved by using nonlinear Mamdani mapping. Additionally, the multi-objective genetic algorithm is employed to find the optimal solutions for a wide-envelope controller covering idle, cruise, and full-power conditions [53]. For system safety constraint handling, the min–max limit protection is improved by using the state feedback method and linear matrix inequalities, which can significantly reduce the probability of transient limit violation [66]. However, due to the vacillating nature of the min–max structure, the stability of the overall system cannot be guaranteed. Thus, a main–max selector scheme with a stability analysis method is further developed by transforming the basic control structure into the form of Lure’s system, where the global asymptotic stability is proved for the control system of a turbofan engine [67]. A novel model-based multivariable limit protection control method is proposed to improve the loss of thrust and is achieved by adaptive command reconstruction, multiple control loop selections, and switch logic [68]. In [69], nonlinear dynamic models of turbomachinery and electrical power systems were developed for the STARC-ABL concept with a partially turboelectric single-aisle propulsion system. A gain-scheduled PI controller with protection logic for turbomachinery and a PI speed controller with a vector controller of pulse width modulation (PWM) was developed based on the piece-wise linear model, and the controller design was examined for the performance metrics of increasing time and operability margins. An optimal control and energy management method for hybrid gas–electric propulsion has been developed, wherein the standard PI controller is adopted to command fuel flow for turbofan speed regulation, and the electromechanical torque is given by linear state feedback and bias, with gains calculated numerically from engine linearization [70].

### 3.2. Hierarchical Control

Hierarchical control is a form of the networked control system, which is widely used for multienergy systems; the main idea is to decompose a complicated control problem into different time-based modules and in turn organize it into layers. This system can be designed in centralized, decentralized, distributed, or hybrid configurations, as shown in Figure 9 [71]. The centralized control model can provide the optimal solution because it has full knowledge of the system and authority over all inputs; however, it is computationally demanding and does not consider the preservation of subsystem privacy. In decentralized control, both subsystems are operated separately by local controllers without communication. This introduces less computational burden than the centralized control strategy. The interactions among subsystems are largely treated as exogenous disturbances to be rejected by subsystem local controllers. Without a consideration of subsystem interactions, the control performance will be degraded. The distributed control can achieve coordinated control actions where local subsystem cost functions are augmented with a global term to account for the impact of their inputs on the entire system [72].

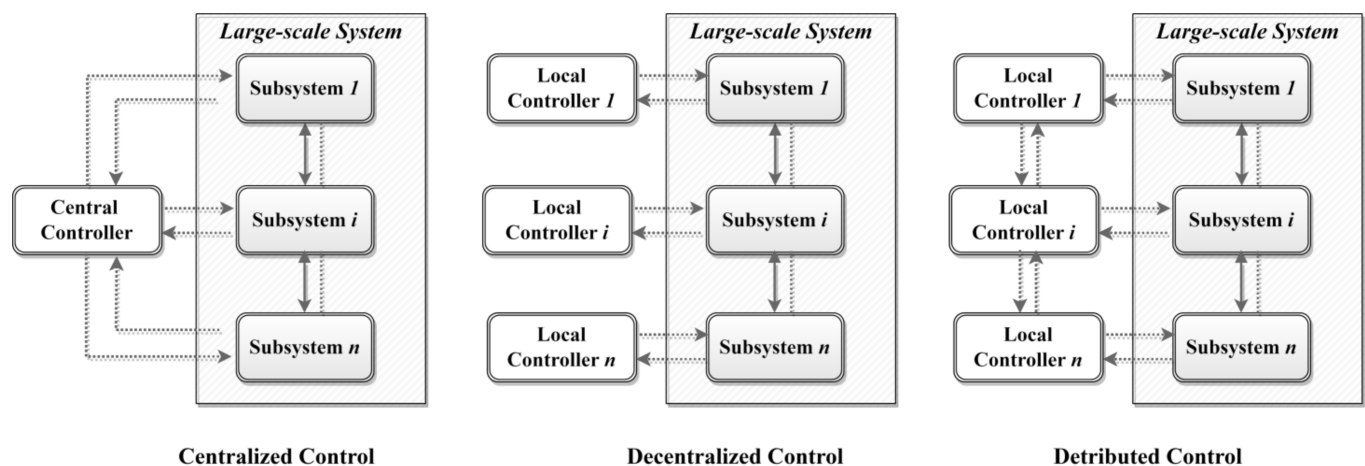


Figure 9. Hierarchical control scheme classification [73].

The hierarchical control scheme is normally implemented with a combination of other control methods, such as classical PI-hierarchical control, hierarchical multiagent control, and hierarchical predictive control. For instance, a two-layer hierarchical architecture is proposed to coordinate the operations of a twin-engine propulsion system, where the upper-propulsion-level controller accounts for load balancing and health management, and each individual engine is operated under the supervision of a local discrete-event supervisory controller at the lower level [74]. Furthermore, a hierarchical multivariable robust control design with nonparallel distributed compensators is proposed for conventional turbofan engines, with the model representation of an uncertain TS fuzzy model, where the level-one compensator ensures the robust performance, and the level-two compensator restrains the uncertainty [75]. Furthermore, a hierarchical model predictive control (MPC) approach is proposed for aircraft electrothermal systems, wherein the coordination of electrothermal systems is achieved by decomposing the multi-energy-domain constrained optimization problem into a set of computationally efficient subproblems that can be solved in real time [76]. For hybrid electric aircraft, a set-based hierarchical MPC framework has been proposed and proven to be a computationally efficient approach to coordinate complex systems across multiple timescales whilst providing guaranteed constraint satisfaction [77]. To face the notable challenge of the coordination of multidomain and multi-timescale system dynamics, a hierarchical control architecture with a predictive controller and graph-based modeling framework has been proposed; the upper-level MPC controller with a long preview horizon is designed to plan the slow timescale system dynamics of battery SOC, the lower-level MPC controller is used to regulate the medium timescale system dynamics of the propeller speed and genset current, and the regulatory embedded controller controls the fastest timescale system dynamics of motor speed and avionic load current directly [78]. A three-level hierarchical load management system has been proposed for other electric aircraft; the upper task level is designed for electrical consumption demand allocation, the medium level is for optimal power distribution, and the lower level is for condition monitoring [79].

### 3.3. Robust Control

Robust control is an approach to controller design that explicitly deals with bounded system uncertainty and disturbances [80]. The most important techniques of robust control are H-infinity control and sliding mode control, which have been extensively used for aero-engine controller design.

For H-infinity control theory, a novel robust gain-scheduling tracking control strategy for aero-engines has been proposed that considers performance degradation. In this study, the onboard engine model with health parameters is represented as piecewise linearized Kalman filters. Then, H-infinity control theory is used for controller design to achieve

good reference tracking with robustness and stability [81]. The H-infinity output tracking problem for aero-engines with a model representation of switched linear-parameter-varying (LPV) systems is also investigated. In [82], a set of switched LPV controllers and a parameter- and mode-dependent switching law are developed to enhance control design flexibility based on multiple-parameter-dependent Lyapunov functions. In [83], the switched LPV model is converted to a switched convex polytopic form with hysteresis switching logic, and then a theoretical sufficiency criterion is provided to guarantee H-infinity performance. Furthermore, a parameter-dependent multiple discontinuous Lyapunov function approach is proposed to solve the H-infinity refined antidisturbance control problem for the switched LPV systems, achieving the required disturbance attenuation and rejection [84]. H-infinity control theory has not been investigated for electrified aircraft propulsion systems to date, but there exist some studies of H-infinity control method applications in hybrid electric vehicle powertrains. In [85], a hierarchical control structure with two robust H-infinity controllers is proposed for plug-in hybrid electric vehicles to achieve a fast and smooth operating mode transition, where the upper-layer H-infinity controller is responsible for calculating the demand clutch torque considering the clutch engaging time and the vehicle jerk, and the lower-layer H-infinity controller aims to perform accurate position tracking control even with the parameter uncertainties and external disturbance. The cooperative control of the drive motor and clutch for the gear shaft is effectively solved by H-infinity loop shaping, even in the presence of model uncertainties and disturbances, as well as the coupling effects between subsystems [86].

Sliding mode control (SMC) is recognized as an efficient approach to robust controller design for complex high-order nonlinear dynamic systems under uncertain conditions. It has the benefits of low sensitivity to system parameter variations and disturbances. Furthermore, SMC can decouple the overall system motion into independent partial subsystems of a lower dimension and consequently reduce the complexity of controller design [87]. In [88], a multi-input and multi-output control scheme is proposed for aero-engine setpoint tracking; it consists of a multivariable SMC controller as a nominal regulator, a series of single-input SMC regulators with min–max selection logic to protect limits, and predictor-based switching logic to ensure the smooth transitions between the main regulator and the limiters. Furthermore, an improved multipower reaching law is proposed for SMC controller design to reduce high-frequency oscillation in aero-engine tracking control [89]. SMC has also been investigated in hybrid electric vehicle powertrain control. For instance, a double-SMC-based control system design is proposed for the efficiency improvement of series hybrid electric vehicles, which also has the advantages of modularity and can be easily implemented on other series hybrid electric vehicle powertrain systems with model and parameter uncertainties [90]. In [91], a fuzzy SMC control method is proposed for the fast and smooth mode transition of parallel hybrid electric vehicles, also offering robustness against vehicle model uncertainties.

### 3.4. Optimal Control

Optimal control is an approach to determine the control signals of a dynamic system to satisfy the physical constraints whilst minimizing the cost function [92]. The optimal control problem is normally formulated as the cost function mapping of system states and control actions, states and input constraints, and system dynamics, which is represented as a set of differential equations with initial conditions [61]. The representative optimal control methods include the linear quadratic regulator (LQR) and the linear quadratic Gaussian (LQG) methods. In LQR control theory, the system dynamics is represented in a linear time-invariant form with the quadratic objective function, and the analytically derived feedback control law is given by using a mathematical optimization algorithm [63]. An LQR-based controller has been developed to stabilize systems and provide better performance in terms of settling time, rising time, and maximum overshoot than a classical PI controller [93]. Additionally, a nonlinear controller was designed using the feedback linearization technique in conjunction with pole placement and LQR theory to guarantee



accurate command tracking and mode blending and ensure the robustness of the engine controller to parameter variations [94]. Furthermore, direct surge margin control for aero-engines has been proposed based on an improved support vector regression (SVR) machine and LQR theory, wherein the SVR machine is used to construct a surge margin prediction model and the LQR controller is improved by augmenting the tracking errors into the state vector to guarantee good robustness and performance over a wide range of power states and flight conditions [95]. For the applications in electrified propulsion systems, an LQR controller has been developed for hybrid electric urban air mobility; this controller can effectively achieve the desired steady state for a wide range of initial conditions [96]. The LQG method has been further developed based on LQR control theory, with a state estimator providing an estimated state vector of measurable outputs [63]. A robust controller for gas turbines has been proposed based on the discrete-time LQG/loop transfer recovery (LTR) approach with self-tuning features, which provides a measure of robustness with regard to both structured uncertainty and the uncertainty caused by the engine operating environment [97]. On this basis, an online estimation method for gas turbine rotation has been further developed based on the LQG/LTR method, which can have only a small effect on the system's natural frequency but allow for a larger open-loop gain to improve the steady state tracking control accuracy [98]. However, this method has not yet been rigorously investigated for applications in electrified aircraft propulsion systems or hybrid electric vehicle powertrains.

### 3.5. Model Predictive Control

Model predictive control (MPC), also known as receding-horizon control, determines the current timeslot control action by solving a finite-horizon optimization problem over the prediction horizon. Then, the prediction horizon is shifted forward, and the optimization calculations are repeated with updated plant states [99]. The MPC method offers advantages when dealing with nonlinearities and handling constraints. However, the closed-loop stability of the MPC control system cannot be guaranteed, and it is inherently less robust than classical feedback, where the optimizations are infeasible with large uncertainties [63,100]. MPC has been extensively applied in the control system design of both conventional propulsion systems and electrified propulsion systems. For example, a multivariable MPC controller was designed and implemented in real time based on a discrete-time linearized state-space model to provide the required thrust and protection limits, where a feedback correction was used to compensate for the model mismatch [101]. In this framework, a semi-alternative-optimization-strategy-based MPC controller was proposed based on the extended Kalman filter turbofan engine model [102]. An advanced constraints management strategy for real-time gas turbine transient control was developed based on the MPC approach. In this study, the MPC controller was designed to identify the engine control modes, which was achieved by using Lagrangian multipliers to handle the constraint inequalities and Hildreth's quadratic programming to select the controller weighting values [103]. To improve the engine response performance and reduce the computational complexity, direct thrust control with nonlinear MPC was proposed based on a linearized online sliding-window deep neural network predictor [104]. For application in EAP systems, a coordinated MPC controller has been developed to achieve the setpoint tracking of large transient thrust and electrical loads whilst ensuring system constraints [56,105]. This study was further developed to improve the match between the rate-based linear prediction model and the nonlinear system by introducing an auxiliary offset state [106]. Furthermore, the bus voltage behaviors were stabilized by imposing a power rate constraint and incorporating this with the energy storage subsystem [107]. Furthermore, a distributed MPC controller was designed based on the alternating direction method of the multiplier. This work presented an innovative approach accounting for the differing control update rates of gas turbine and electrical powertrain subsystem local controllers and the preservation of subsystem privacy [108].

**Table 9.** Overall characteristics of classical control theory and modern control theory [61].

Methodology	Benefits and Limitations	Applications
<u>Classical Control</u> <i>PID controller</i>	Simple implementation in practice with relative efficiency. Only applied in single-input–single-output (SISO) systems, requires the system model to be linear-time-invariant (LTI), cannot incorporate constraints, and lacks overall performance optimization.	Conventional propulsion system: [49,52,55,64–68] Electrified propulsion system: [69,70]
<u>Hierarchical Control</u> <i>Centralized control, decentralized control, distributed control</i>	A form of networked control system; a computationally efficient approach for multidomain, multi-timescale system dynamics; decomposes a complicated control problem into different time-based modules and, in turn, organizes them into layers.	Conventional propulsion system: [74–76] Electrified propulsion system: [77–79]
<u>Robust Control</u> <i>H-infinity, sliding mode control</i>	Explicitly deals with bounded system uncertainty and disturbances. Decouples the overall system motion into independent partial subsystems of lower dimension and consequently reduces the complexity of feedback design.	Conventional propulsion system: [81–84,87–89] Electrified propulsion system: [85,86,90,91]
<u>Optimal Control</u> <i>LQR, LQG, <math>H_2</math></i>	Determines the control signals of a dynamic system to satisfy the physical constraints whilst minimizing the cost function.	Conventional propulsion system: [93–95,97,98] Electrified propulsion system: [96]
<u>Model Predictive Control</u>	Flexible nonlinearity and constraint handling; closed-loop stability cannot be guaranteed; poor robustness.	Conventional propulsion system: [101–104] Electrified propulsion system: [59,105–107]

**Table 10.** Summary of aircraft propulsion system control methodologies [61].

	Classical Control Theory	Advanced Control Theory
Domain	Frequency, S-domain	Time domain
Model representation	Transfer function	State-space
Continuity	Continuous	Continuous, discrete, hybrid
Linearity	Linear	Linear, nonlinear
Time variance	Time-invariant	Time-variant
Dimensions	Single input, single output	Multiple input, multiple output
Determinism	Deterministic	Deterministic, stochastic
Implementation	Cheap, easy	Expensive, complex

#### 4. Electrified Aircraft Propulsion System Supervisory Energy Management Strategy

Energy management strategies (EMSs) have been extensively explored for hybrid electric vehicles, microgrids, and power grids [108,109]. However, HEA EMS consideration is recognizably different from these ground-based energy systems. Especially for commercial aviation, the mission analysis and performance assessment are based on the premise of balancing the thrust to drag with the fuel flow and electrical power flow within the set operational and performance constraints defined by the top-level requirements. Normally, the mission segments are discretized by an energy-based approach with a mission explicitly defined for each step, which involves a large number of iterations between the propulsion system performance module and the aircraft performance module based on

a multidisciplinary analysis of the aerodynamics, propulsion system performance, and weight property [110]. Moreover, the weight is a major consideration for flight mission analysis, especially for hybrid electric aircraft, since the additional weight of the electrical powertrain and battery energy storage will degrade the integrated aircraft performance due to the induced weight-related drag penalty [111]. Propulsion system electrification and hybridization can ultimately improve block fuel burn and emissions at the isolated propulsion system level due to electrical power on-take and more efficient operation, but the promising improvements and benefits cannot be guaranteed from the perspective of the integrated aircraft level, due to the weight-related drag penalty [36]. Furthermore, the in-flight gas turbine fuel flow also leads to changes in aircraft mass and further affects the aerodynamics performance, such as lift-dependent drag and thrust demand, especially for regional jets that operate at a high altitude with long flight durations. Therefore, a fully coupled propulsion–aerodynamic model with multidisciplinary analysis is required for hybrid electric propulsion system performance assessment at the integrated aircraft level. Furthermore, energy onboard should be equal to mission energy demand as a key system constraint, since the carrying of an excessive amount of energy induces a weight and drag penalty [112]. For hybrid electric aircraft with dual energy sources, both the consumable mass energy sources of the fuel and the nonconsumable mass energy sources of the battery should be considered. Thus, hybrid electric aircraft EMSs should be considered from a global optimization perspective at the integrated aircraft level with a fully coupled aerodynamics–propulsion multidisciplinary model, which will also greatly increase the system complexity and computational load. Thus, the HEA EMS is complicated, challenging to implement, and has not yet been widely investigated.

This section provides a comprehensive survey of the HEA EMS literature to date, systematically summarizes the state-of-the-art HEA EMS technology, and explores the research trends in this field. EMSs can be classified as rule-based and model-based optimization methods. Model-based optimization strategies can be further divided into global optimization EMSs and instantaneous optimization EMSs, as shown in Table 11.

**Table 11.** Summary of energy management strategy methodologies.

Methodologies	Advantages	Disadvantages
<u>Rule-Based EMS</u> <i>Deterministic rule-based EMS,</i> <i>fuzzy rule-based EMS.</i>	Easy to implement, low computational load.	Aircraft performance is determined by predefined rules; highly dependent on designers' expertise; highly sensitive to flight mission profile.
<u>Global Optimization EMS</u> <i>Convex programming (CP),</i> <i>dynamic programming (DP),</i> <i>Pontryagin's minimum principle (PMP);</i> <i>metaheuristic algorithms: genetic algorithm (GA),</i> <i>particle swarm optimization (PSO), differential</i> <i>evolution algorithm (DEA).</i>	Provide a globally optimal solution (all); stochastic solution generation to avoid local optima (metaheuristic algorithms).	Analytical methods are frequently not applicable for complicated problems with complex constraints (PMP); strong model simplification (CP, DP); require a planned flight mission profile as a priori knowledge (DP, metaheuristic algorithms); high computation effort (metaheuristic algorithms).
<u>Instantaneous Optimization EMS</u> <i>Equivalent consumption minimization strategy</i> <i>(ECMS),</i> <i>model predictive control (MPC).</i>	Easy to implement (ECMS); reduced computation load; allow the current timeslot to be optimized while keeping account of future timeslots; handle many system constraints simultaneously; applicable to a multivariable problem (energy balance of multiple sources of energy); applicable to a multiobjective optimization problem (MPC).	Achieves optimal instantaneous equivalent fuel consumption but cannot guarantee the optimal aircraft performance at mission level (ECMS); highly sensitive to flight mission profile (ECMS); single-objective optimization, cannot be expanded to operating costs, emissions, etc. (ECMS).

#### 4.1. Rule-Based EMS

The rule-based EMS is an effective approach for real-time implementation; it relies on a set of predefined rules to take control actions instead of involving explicit minimization or optimization [113]. The design of rules is generally based on heuristics, designers' expertise, or the insights of global optimal solutions through the offline optimization process. Thus, the rule-based EMS is practical to implement and has a low computational load, but the

HEA performance is highly dependent on the designers' expertise and sensitive to the flight mission profile.

A modeling approach for the rule-based EMS was investigated for the AirTaxi mission of hybrid electric rotorcraft to reduce fuel consumption. It proposed and compared four modes with the consideration of battery aging effects: only thermal engine operation, only electric machine (EM) operation as a motor, both engine and EM operation with battery discharging, and both engine and EM operation with battery recharging [114]. In [115], a rule-based controller is designed to manage the power flow of the parallel HEPS with three operating modes: internal combustion engine (ICE) only, ICE and EM, and ICE and GE. By the utilization of both electric machines and internal combustion engines, HEPS can effectively achieve fuel saving and a reduction in take-off noise and emission levels. The systematic hybrid aircraft power schedule optimizer (SHAPSO) is proposed in [116]. Additionally, a rule-based EMS is presented to determine the optimal power management of a parallel HEPS with the goals of mission block fuel burn reduction over the flight envelope. The rule-based power schedules of constant power, power at climb, and power at cruise are proposed and compared with the optimal control method and dynamic programming. A fuzzy state machine (FSM)-based EMS has been proposed to control the power flow for a hybrid electric UAV for online application [117]. It innovatively couples fuzzy logic with a state machine, which can satisfy the power demand and effectively achieve block fuel reduction for real-time implementation. A fuzzy logic parameter updating (FLPU) method was developed to achieve adaptive power distribution for a light electric aircraft with the dual energy sources of battery energy storage and fuel cells. The proposed method could adaptively update the power distribution algorithm based on the aircraft operation conditions [118].

#### 4.2. Convex Programming

Convex programming (CP) is widely used to solve optimal control problems in aerospace engineering for onboard real-time applications because of its fast convergence and global optima [119]. A general convex optimization problem can be described as Equation (1):

$$\begin{aligned} & \text{Minimize } f_0(x) \\ & \text{subject to } f_i(x) \leq b_i, \quad i = 1, \dots, m, \end{aligned} \quad (1)$$

where the functions  $f_0, \dots, f_m : R^n \rightarrow R$  are convex, i.e., satisfy

$$f_i(\alpha x + \beta y) \leq \alpha f_i(x) + \beta f_i(y) \quad (2)$$

for all  $x, y \in R^n$  and all  $\alpha, \beta \in R$  with  $\alpha + \beta = 1$ ,  $\alpha \geq 0, \beta \geq 0$ .

The algorithms are only applicable when the problem is strictly formulated in convex terms: (1) the objective function must be convex, (2) the inequality constraint functions must be convex, and (3) the equality constraint functions must be affine [120]. Thus, the EMS using convex optimization requires strong model simplifications to transform a nonconvex problem into a convex problem. For instance, HEA optimal energy management is formulated as a convex problem to minimize the fuel consumed over a planned flight path, where the point-mass aircraft dynamic model, the electrical powertrain, and the gas turbine are simplified using quadratic approximations. Furthermore, this study requires the drive power to be approximated a priori by assuming constant aircraft mass for the duration of the flight [121]. A convex multiobjective optimization method is proposed for a hybrid electric aircraft to minimize fuel consumption and polluted emissions during the entire flight mission. The optimal EMS problem is formulated as mixed-integer nonlinear programming by replacing the logic constraints of battery energy storage system operation with the Karush–Kuhn–Tucker optimality condition [122]. A real-time supervisory energy management control method is proposed for HEA series and parallel propulsion systems. The convex formulation is derived from four simplifications: (1) a prior flight mission profile and power demand; (2) convex propulsion system loss map functions based on

an assumption of the monotonicity of the loss map; (3) reformulating the point-mass aircraft dynamics as a quadratic function; and (4) relaxing the power balance equality to the inequality. This convex optimization problem is solved by the alternating direction method of multipliers (ADMM) algorithm, which can reduce the computation time by orders of magnitude compared with the general convex interior point solver, CVX [123]. In summary, the challenge of using convex optimization lies in recognizing and formulating the convex optimization problems. Once a problem is formulated as a convex optimization problem, it can be solved efficiently using algorithms, e.g., the gradient descent method, the steepest descent method for unconstrained minimization problems, Newton's method for equality-constrained minimization problems, and interior-point methods for inequality-constrained minimization problems [120].

#### 4.3. Dynamic Programming

Dynamic programming (DP) is a model-based numerical optimization method based on Bellman's principle of optimality. The DP algorithm is widely used in energy management strategy problems to provide the global optimal solution [124]. In the DP algorithm, the problem is gridded according to time, states, and control signals and then divided into a sequence of subproblems which are solved recursively backward in time from the final step to the first. The optimal solution is achieved by minimizing the cost function, which is defined as Equation (3) [108]:

$$J = \sum_{k=0}^{N-1} g_k(x_k, u_k, w_k) + g_N(x_N) \quad (3)$$

where  $g_N$  represents the terminal cost;  $g_k$  is the cost function incurred at the time slot  $k$ ; and  $x_k, u_k, w_k$  denote system states, control decision, and disturbances, respectively, at the time slot  $k$ .

The cost function of each DP subproblem is defined as Equation (4):

$$J_k(x_k) = \min_{u_k \in U_k(x_k)} \left\{ \underbrace{g_k(x_k, u_k, w_k)}_{\text{stage cost}} + \underbrace{J_{k+1}(f_k(x_k, u_k, w_k))}_{\text{cost-to-go}} \right\} \quad (4)$$

where  $k = N - 1, N - 2, \dots, 0$  and the optimal solution is calculated backwards from  $N-1$  to 0 with the initialization of terminal cost  $J_N = g_N(x_N)$ . The DP algorithm requires the decomposition of the problem into subproblems, so the optimal solution accuracy highly depends on the resolutions of the problem discretization. Additionally, the computational demand will exponentially increase with the number of gridded variables, which is called "the curse of dimensionality" [125].

The application of the DP algorithm in hybrid electric aircraft was first investigated to optimize the power management and torque split of the powertrain for a parallel hybrid electric UAV considering terminal state constraints. The study proposed a quasi-real-time iterative algorithm based on the DP algorithm and further explored new flight capabilities by the utilization of hybrid powertrain architecture [126]. A DP-algorithm-based EMS was proposed to find an optimal power-splitting sequence to minimize the fuel consumption over the whole flight mission for a hybrid electric lightweight rotorcraft. The optimal EMS control problem was modeled as a graph structure, where the splitting sequences were represented as the paths with the transition cost weights. Based on the graph structure model, the optimal splitting sequence (shortest energy path) was obtained by using the Dijkstra algorithm [127]. Furthermore, an offline EMS optimization method was proposed using DP to minimize the fuel consumption in a hybrid electric helicopter for urban air-mobility considering the effect of battery aging. This study assumed a planned flight mission profile and considered a constant take-off weight, with the saved fuel being converted into payload [128]. Similarly, an optimal energy management methodology was



proposed to minimize fuel consumption based on the DP algorithm with a prescribed flight mission; a point of originality was that this study took into account energy recovery via a propeller during descent [129]. However, these studies assumed a fixed required propulsive power demand as prior knowledge, failing to consider the propulsion–aerodynamic coupling effects and neglecting the variation in aircraft weight during flight. In summary, the DP algorithm can provide the optimal global solution, but it is not applicable for real-time implementation because it requires a priori information and a backward discretized model [125]. Therefore, the DP algorithm is normally used to design rules for rule-based EMSs or as a benchmark to evaluate other EMS solutions.

#### 4.4. Pontryagin's Minimum Principle

Pontryagin's minimum principle (PMP) is a model-based analytical optimization method that can provide a globally optimal solution by transforming a global optimization problem described by DP into an instantaneous Hamiltonian optimization problem and then minimizing the Hamiltonian function according to the co-state [130]. This method has been widely used in the EMS problem of hybrid electric vehicles [131–133]; however, few studies have investigated PMP method application in the field of HEA EMS problems. This is because analytical solutions are frequently not applicable for complicated problems with complex constraints [134].

The Hamiltonian function is defined as Equation (5) [113]:

$$H(x(t), u(t), \lambda(t), t) = L(x(t), u(t), t) + \lambda(t)^T \cdot f(x(t), u(t), t) \quad (5)$$

where  $f(x(t), u(t), t)$  is the system dynamics model;  $L(x(t), u(t), t)$  is the instantaneous cost; and  $\lambda(t)$  is a vector of the optimization variables—*co-states* of the system. In PMP, the following conditions should be satisfied if  $x^*(t)$  is the optimal control law.

1. The state and co-state should satisfy the following conditions, as represented by Equations (6)–(9):

$$\dot{x}^*(t) = \left. \frac{\partial H}{\partial \lambda} \right|_{u^*(t)} = f(x^*(t), u^*(t), t) \quad (6)$$

$$\begin{aligned} \dot{\lambda}^*(t) &= - \left. \frac{\partial H}{\partial x} \right|_{u^*(t)} = h(x^*(t), u^*(t), \lambda^*(t), t) \\ &= - \frac{\partial L}{\partial x}(x^*(t), u^*(t), t) - \lambda^*(t) \cdot \left[ \frac{\partial f}{\partial x}(x^*(t), u^*(t), t) \right]^T \end{aligned} \quad (7)$$

$$x^*(t_0) = x_0 \quad (8)$$

$$x^*(t_f) = x_{\text{target}} \quad (9)$$

2. For all  $t \in [t_0, t_f]$ ,  $u^*(t)$  globally minimizes the Hamiltonian function:

$$\begin{aligned} H(u(t), x^*(t), \lambda^*(t), t) &\geq H(u^*(t), x^*(t), \lambda^*(t), t), \forall u(t) \in U(t), \forall t \in [t_0, t_f] \\ \text{i.e., } u^*(t) &= \underset{u(t) \in U(t)}{\operatorname{argmin}} H(u(t), x(t), \lambda(t), t) \end{aligned} \quad (10)$$

where  $U(t)$  is the set of admissible control values at time  $t$ . This optimal control problem using the minimum principle is normally solved by the shooting method. This method applies an arbitrary initial value of the co-state at the beginning,  $\lambda_0$ , and then solves the minimization problem for the dynamic model at each time slot. The optimization procedure ends, and the optimal solutions can be obtained if the final values of the state and co-state can satisfy the terminal constraints. Otherwise, the initial value of  $\lambda$  should be changed until the solution meets all the constraints.

In the research field of HEA EMSs, a PMP-based real-time EMS has been innovatively proposed for hybrid electric aircraft propulsion systems to achieve emission reductions and system efficiency improvements. By applying PMP, the EMS optimal control problem with constraints is converted into a point-wise optimization with the battery pack temperature and SOC as co-states to achieve a computationally efficient solution. Additionally, the formulation of the EMS optimal control problem using PMP strongly depends on model simplification, e.g., assuming a priori knowledge of mission profile data and thrust request [135].

#### 4.5. Metaheuristic Algorithms

Metaheuristic algorithms are derivative-free methods inspired by nature, which are generally utilized to solve optimization problems with a large search space of likely solutions, e.g., simulated annealing (SA), genetic algorithm (GA), differential evolution algorithm (DEA), ant colony optimization (ACO), particle swarm optimization (PSO), and cuckoo search (CS) [108]. These metaheuristic algorithms employ alternative methods to populate candidates for an optimal solution instead of derivative calculations, and the optimization performance and convergence behavior highly depend on the selection of hyperparameters to avoid local minima. These methods have been extensively investigated in offline HEA EMS optimal control problems.

A representative nature-inspired method, the PSO algorithm is inspired by social organism movements in groups, such as swarms and ant colonies. This algorithm populates the states, positions, and velocities of the particles. These particles can interact locally with each other to share information and then store their best positions and the group's best solution to improve the next population until the target is achieved [136]. The PSO algorithm can be expressed as Equations (11) and (12) [137]:

$$v_d^i(k+1) = wv_d^i(k) + c_1 \cdot r_1 \cdot (pBest_i(k) - x_d^i(k)) + \dots + c_2 \cdot r_2 \cdot (gBest_i(k) - x_d^i(k)) \quad (11)$$

$$x_d^i(k+1) = x_d^i(k) + v_d^i(k+1) \quad (12)$$

where  $i$  represents each particle;  $pBest$  and  $gBest$  are particle best position and global best position; and  $w$ ,  $c_1$ ,  $c_2$ ,  $r_1$ ,  $r_2$  are inertia weight, positive parameters, and random parameters within [0,1], respectively. Maximum and minimum particle velocity should be within  $\pm v_d^{\max}$ . The PSO algorithm can be effectively applied in the optimization problem with complex objective functions, where only two variables (particle position and velocity) are required in population and iteration.

Another typical metaheuristic optimization method is the GA, which is inspired by natural evolution with the procedures of reproduction, crossover, and mutation and involves randomness to guarantee population diversity [138]. As shown in Figure 10, the algorithm creates a population of individual solutions (chromosome) and randomly selects individuals from the current population as parents to produce the children for the next generation at each generation. Over successive generations, the population converges to an optimal solution where a balance of computational load and precision is achieved [139]. Following the general outline of the GA, the nondominated sorting genetic algorithm II (NSGA II) has been developed to solve a Pareto-optimal front for the constrained multiobjective optimization problem [140].

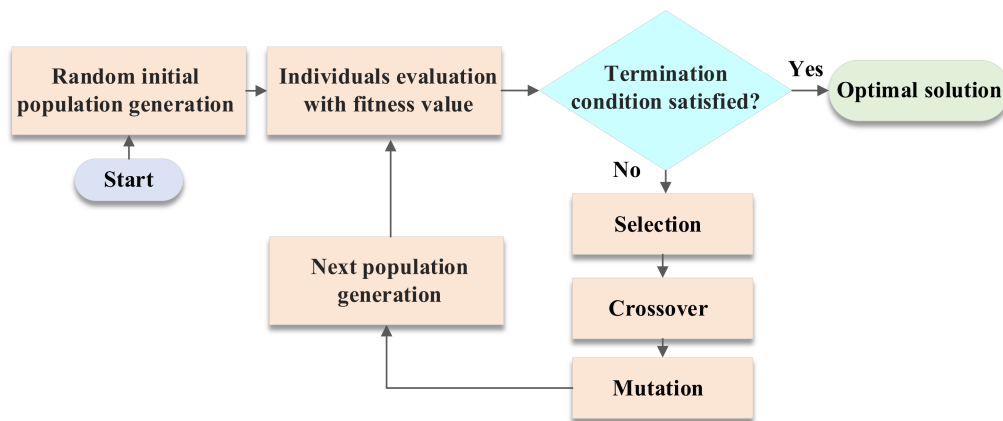


Figure 10. Flow chart of genetic algorithm.

Metaheuristic algorithms have been extensively used in energy management for electric aircraft. A comparative analysis of the most efficient up-to-date metaheuristic algorithms, such as the artificial bee colony (ABC), grey wolf optimization (GWO), the cuckoo search (CS), and the moth swarm algorithm (MSA), has been presented in the literature [141]. It evaluates the performance of metaheuristic algorithms in the application of energy management for fuel-cell-based hybrid emergency power units in electrical aircraft. In another study [142], the synergy effects between powertrain, structure, and mission are exploited by applying the multiobjective optimization method, NSGA-II, and the S-metric selection evolutionary multiobjective algorithm (SMS-EMOA), and the potential benefit of structural integration and multifunctionalization are addressed. The nonlinear nature of the HEPS for aircraft is highlighted in [143], and DEA techniques are applied for power management in series-, full-electric-, and turboelectric-powered UAVs, respectively. An active energy management strategy is proposed for a fuel-cell/battery/supercapacitor-powered aircraft to control the battery and supercapacitor SOC as well as to minimize hydrogen consumption by applying metaheuristic algorithms including the ant lion optimizer (ALO), moth-flame optimization (MFO), the dragonfly algorithm (DA), the sine cosine algorithm (SCA), the multi-verse optimizer (MVO), particle swarm optimization (PSO), and the whale optimization algorithm (WOA) [144]. Rendón et al., in [145], proposed a model-based fully coupled propulsion–aerodynamics optimization framework for the HEA EMS problem by using the NSGA-II algorithm to address the design trade-off, i.e., block fuel burn, energy consumption, and emissions. This study showcased an innovative approach considering the sizing cascading effects in HEA propulsion system design and mission analysis.

#### 4.6. Equivalent Consumption Minimization Strategy

The equivalent consumption minimization strategy (ECMS) is a heuristic method to provide effective solutions for optimal control problems. The ECMS algorithm introduces an *electricity-to-fuel equivalent factor*, which converts electricity consumption into fuel consumption to allow for unifying fuel and electricity consumption in a single objective. The instantaneous equivalent fuel consumption is minimized to obtain instantaneous optimal power distribution, which is defined as Equations (13) and (14), see [113,146]:

$$\dot{W}_{\text{feq}}(t) = \dot{W}_f(t) + Q(t) \cdot \frac{P_{\text{Batt}}(t)}{LHV} \quad (13)$$

$$\begin{aligned} \text{Global : } \min_{P_{\text{Batt}}(t) \in U_{\text{Batt}}} \int_{t_0}^{t_f} \dot{W}_f(t) dt, \quad \text{SOC}_{\min} < \text{SOC} < \text{SOC}_{\max} \\ \text{Instantaneous : } \int_{t_0}^{t_f} \min_{P_{\text{Batt}}(t) \in U_{\text{Batt}}} \dot{W}_{\text{feq}}(t) dt, \quad \text{SOC}_{\min} < \text{SOC} < \text{SOC}_{\max} \end{aligned} \quad (14)$$

where  $\dot{W}_f(t)$ ,  $Q(t)$ ,  $P_{Batt}(t)$ ,  $\dot{W}_{feq}(t)$ ,  $LHV$ , and  $t$  are actual fuel flow rate, electricity-to-fuel equivalent factor, battery power, equivalent fuel consumption, fuel lower heating value, and time step, respectively.

Using the ECMS method, the global optimization problem is converted to a local instantaneous optimal problem. Due to this, the ECMS algorithm is a more computationally efficient and easier-to-implement method that is suitable for online applications. It can provide control actions close to globally optimal solutions with the tuning of the electricity-to-fuel equivalent factor. The optimal equivalent factor is obtained by offline optimization; thus, the aircraft performance of the ECMS is highly sensitive to the flight mission profile.

A fuzzy-logic-based equivalent consumption optimization method has been proposed for hybrid-electric-powered unmanned aerial vehicles (UAVs) [147], which achieves the suboptimal hybridization of fuel and electrical energy for online optimization and sustains the battery SOC within the desirable range. Furthermore, the authors of [36] developed an ECMS with hard constraints and an ECMS with a penalty function and compared them with a global optimization (GA)-based EMS. With the offline tuning of the equivalent factor, ECMS methods can effectively obtain similar optimal solutions to global optimization (GA)-based EMS methods but significantly reduce the computational time. In [128], an online optimizer was developed based on the knowledge of global optimal solutions provided by dynamic programming, which took particular account of the desired SOC and the actual state of health (SOH) and could be opportunely adapted to specific cases.

#### 4.7. Model Predictive Control

Model predictive control (MPC) has been utilized in an online EMS optimization algorithm to obtain the optimal control actions, where future information is incorporated into the control problem to improve EMS performance [36]. At each sampling time, the algorithm uses the system model to predict the future states of the plant over the prediction horizon with length  $N$ , and then the optimal control policy from  $t$  to  $t+N$  is obtained by rolling optimization. After that, the control policy of the current sampling time is applied regardless of the rest, and the system state is updated accordingly. The MPC-based EMS performance is determined by model fidelity, sampling step, and the length of the prediction horizon [1]. The length of the prediction horizon is tuned accordingly to consider the trade-off between the computational effort and optimization performance. This allows the current timeslot to be optimized while keeping future timeslots in mind. Thus, the MPC EMS can significantly reduce the computational effort as compared to the global optimization algorithm while achieving very similar optimization performance [36].

A nonlinear MPC-based EMS was proposed for parallel HEPS to minimize mission block fuel burn, energy consumption, and emissions, based on a fully coupled aerodynamics-propulsion HEA model. Additionally, the penalty function was introduced to reduce the computational effort but effectively maintain the system constraints [36]. Furthermore, a system-level centralized optimal load and energy management strategy was proposed for aircraft microgrids to schedule the battery system, provide flexibility in the demand side, and satisfy time-varying operational requirements. This was achieved by using a multiobjective MPC model with a mixed-integer linear programming algorithm (MILP) [147]. A predictive control strategy was proposed for real-time implementation that took into account the variation in aircraft mass during flight. At each sampling interval, rolling optimization was achieved by a convex formulation of the problem of minimizing fuel consumption over a predicted future flight path [121,148].

## 5. Summary

Electrified aircraft propulsion technologies, forming a very promising approach to sustainable aviation decarbonization, were investigated and explored. In particular, this paper provided comprehensive insights into the recent progress in control system design and energy management aspects for electrical aircraft propulsion systems, together with a detailed analysis of the emerging challenges and technical barriers in this new and chal-

lenging research field. The control design of aircraft propulsion is a rather demanding task, while the operation of gas turbines is an extremely complex nonequilibrium and nonlinear aerothermodynamic process. In addition to this, a set of performance and safety requirements regarding the steady state accuracy, transient accuracy, disturbance rejection, stability, stall margin, structural integrity, and engine degradation must be fulfilled. For electrified aircraft propulsion, multiple challenges and technical barriers have emerged: (1) the coupling and interactions between the gas turbine and electrical powertrain, involving mechanical, hydraulic, thermal, electrical, and electronic elements; (2) the negative impedance instability behavior of the power load regulation of the electrical powertrain subsystem; (3) multiple timescales between the gas turbine and electrical powertrain subsystems; and (4) the more frequent failure of the EAP system due to the diversity of components and the coupled operation. Energy management strategies have been extensively explored in the literature for hybrid electric vehicles, microgrids, and power grids; however, EMSs for aircraft/aerial systems are undoubtedly more challenging compared to the ground-based energy systems mentioned above. Especially for commercial aviation, the mission analysis and performance assessment are based on balancing the thrust to drag with the fuel flow and electrical power flow based on a multidisciplinary analysis of the aerodynamics, propulsion system performance, and weight property. In addition, aircraft weight is a significant concern for flight mission analysis. The additional weight of the electrical powertrain and battery energy storage will degrade the aircraft performance due to the induced weight-related drag penalty. Thus, a fully coupled propulsion–aerodynamic model with multidisciplinary analysis is required for hybrid electric propulsion system performance assessment at the integrated aircraft level. While this topic has received increased attention in recent years, there are significant challenges to enabling commercially available sustainable electrification in the aviation sector.

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## Abbreviations

HEA	Hybrid Electric Aircraft
EM	Electric Machine
GE	Generator
MTOW	Maximum Take-Off Weight
SE	Specific Energy
SP	Specific Power
PAX	Passenger
YEIS	Year of Entry Into Service
MIPH	Mechanically Integrated Parallel Hybrid
CIPH	Cycle-Integrated Parallel hybrid
DC	Direct Current
EAP	Electrified Aircraft Propulsion
EEC	Electronic Engine Control
VAFN	Variable-Area Fan Nozzle
VBV	Variable Bleed Valve
PID	Proportional Integral Derivative
SISO	Single Input Single Output



RBF	Radial Basis Function
PWM	Pulse Width Modulation
MPC	Model Predictive Control
LPV	Linear Parameter Varying
SMC	Sliding Mode Control
LQR	Linear Quadratic Regulator
LQG	Linear Quadratic Gaussian
LTR	Loop Transfer Recovery
EMSs	Energy Management Strategies
CP	Convex Programming
DP	Dynamic programming
PMP	Pontryagin's Minimum Principle
GA	Genetic Algorithm
PSO	Particle Swarm Optimization
DEA	Differential Evolution Algorithm
ECMS	Equivalent Consumption Minimization Strategy
SHAPSO	Systematic Hybrid Aircraft Power Schedule Optimizer
FSM	Fuzzy State Machine
FLPU	Fuzzy Logic Parameter Updating
ADMM	Alternating Direction Method of Multipliers
SA	Simulated Annealing
ACO	Ant Colony Optimization
CS	Cuckoo Search
NSGA II	Nondominated Sorting Genetic Algorithm II
ABC	Artificial Bee Colony
GWO	Grey Wolf Optimization
MSA	Moth Swarm Algorithm
SMS-EMOA	S-Metric Selection Evolutionary Multi-Objective Algorithm
ALO	Ant Lion Optimizer
AFO	Moth Flame Optimization
DA	Dragonfly Algorithm
SCA	Sine Cosine Algorithm
MVO	Multi-Verse Optimizer
WOA	Whale Optimization Algorithm
UAVs	Unmanned Aerial Vehicles
SOH	State of Health
MILP	Mixed-Integer Linear Programming Algorithm

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